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Additional Information

2 **ASSESSMENT OF COMPRESSIBILITY EFFECTS ON INTERNAL NOZZLE**
3 **FLOW IN DIESEL INJECTORS AT VERY HIGH INJECTION PRESSURES**

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14 **ABSTRACT**

15 Diesel fuel injection systems are being used at higher injection pressure conditions over
16 time because of more stringent emissions requirements. Thus, the importance to properly
17 take into account the fluid compressibility on injection CFD simulations is also
18 increasing. In this paper, an investigation of the compressibility effects in nozzle flow
19 simulations has been carried out for injection pressures up to 250 MPa. To do so, the fluid
20 properties (including density, viscosity and speed of sound) have been measured in a wide
21 range of boundary conditions. These measurements have allowed to obtain correlations
22 for the fluid properties as a function of pressure and temperature. Then, these equations
23 have been incorporated to a CFD solver to take into account the variation of the fluid
24 properties with the pressure changes along the computational domain. The results from

25 these simulations have been compared to experimental mass flow rate and momentum
26 flux results, showing a significant increase in accuracy with respect to an incompressible
27 flow solution.

28

29 **KEYWORDS:** nozzle, modelling, Diesel, dynamic, compressibility, CFD

30 **NOMENCLATURE**

a_f Fuel speed of sound

A_o Geometrical nozzle outlet area

C_d Discharge coefficient, $C_d = \frac{\dot{m}}{\rho_f \cdot A_o \cdot u_b}$

D_i Geometrical nozzle inlet diameter

D_o Geometrical nozzle outlet diameter

$k\text{-factor}$ Nozzle conicity, $k\text{-factor} = \frac{D_i[\mu m] - D_o[\mu m]}{10}$

\dot{m} Mass flow

\dot{M} Momentum flux,

P Fluid pressure

P_b Discharge pressure

P_{inj} Injection pressure

T Fluid temperature

u_{eff} Effective outlet nozzle orifice velocity

u_b Theoretical outlet orifice velocity, $u_b = \sqrt{\frac{2 \cdot (P_{inj} - P_b)}{\rho_f}}$

Greek Symbols

ΔP Pressure drop, $\Delta P = P_{inj} - P_b$

ρ_f Fuel density

ν_f Kinematic viscosity

μ_f Dynamic viscosity

μ_0 Dynamic viscosity at 0.1MPa pressure

31

32 **1. INTRODUCTION.**

33 In the last decades, diesel engine researchers have focused on minimizing the exhaust
34 emissions maintaining the thermal efficiency advantage compared to gasoline engines. In
35 particular, efforts have been made to achieve a combined reduction of nitrogen oxides
36 and soot particles, which are characteristic of the lean diffusive combustion process
37 existing in such engines [1], [2].

38 Two main paths have been followed to reduce exhaust emissions in diesel engines. On
39 the one hand, several aftertreatment components, such as Diesel Particulate Filter (DPF),
40 Diesel Oxidation Catalyst (DOC), Selective Catalyst Reduction (SCR) or Lean-NOx Trap
41 (LNT) have been placed at the engine outlet to collect and/or convert the exhaust
42 emissions before reaching the atmosphere [3], [4]. On the other hand, new combustion
43 modes with high levels of Exhaust Gas Recirculation (EGR) and higher rates of premixed

44 combustion have been implemented to reduce the emissions at engine-out [5]–[8]. The
45 performance of the fuel injection system has been proven as critical for such strategies,
46 since it controls the atomization and fuel-air mixing processes [9]–[11].

47 Many authors have tried to study the characteristics of the flow inside the fuel injector,
48 and in particular inside the nozzle orifices. Several studies have made use of transparent
49 geometries for this purpose, but many of them explored simplified geometries [12]–[15]
50 or were significantly limited in the maximum achievable injection pressure [16]–[18].
51 Thus, Computational Fluid Dynamics (CFD) tools have been developed on the last
52 decades as a tool to get further insight in the relationship between the nozzle geometry,
53 the internal flow characteristics and the hydraulic conditions at the nozzle exit [19]–[22],
54 which are a necessary input for spray combustion models [23]–[26].

55 The fuel physical properties (mainly density and viscosity) have a significant impact on
56 the internal nozzle flow characteristics. Battistoni et al [23] compared the internal flow
57 and near-nozzle spray details for a standard diesel fuel and a soybean methyl ester (SME),
58 showing that the different viscosity among them severely impacts both the outlet mass
59 flow rate and the spray features. Similar conclusions about the effect of the fuel properties
60 have already been seen both experimentally and numerically for other kinds of biodiesel
61 [27]–[31] and for winter fuel formulations [32]–[34]. Recently, a few authors [35]–[39]
62 have showed that it is important to consider not only the changes in the fuel properties
63 related to the fuel composition, but also those related to the different temperature and
64 pressure conditions along the nozzle geometry, which are traditionally neglected.

65 In the current paper, an effort to understand the impact of compressibility effects on
66 internal nozzle flow simulations at very high injection pressure (up to 250 MPa) has been
67 performed. For this purpose, the fuel used for the study has been widely characterized at

68 different levels of temperature and pressure, producing the corresponding correlations for
69 the fuel density, viscosity and speed of sound. Then, the hydraulic behavior of the injector
70 has been determined in terms of injection rate and momentum flux for different levels of
71 injection pressure and backpressure. These results have been finally compared to internal
72 flow CFD simulations carried out with two strategies: constant fuel properties
73 (incompressible) and pressure-dependent fuel properties (compressible). This procedure
74 allows to quantify the differences obtained in the main flow parameters when
75 compressibility effects are considered compared to the more simple incompressible
76 solution generally seen in the literature [15], [40], [41].

77 The paper is structured in 5 sections. In section 2, the main experimental methodologies
78 used along the study are described, together with the correlations obtained for the main
79 fuel physical properties. Section 3 details the setup used for the internal flow CFD study,
80 whose main results are depicted in Section 4. Finally, the main conclusions obtained from
81 the work are drawn in Section 5.

82 **2. EXPERIMENTAL TOOLS**

83 In this section, the main experimental techniques used for the study are briefly described.

84 **2.1. Nozzle geometry determination**

85 For the current study, a solenoid-driven diesel injector with a 5-orifices convergent nozzle
86 has been used. In order to perform the internal nozzle flow simulations, it is necessary to
87 have all its geometrical details. To do so, a previously developed and validated silicone
88 molding technique has been employed. The technique is based on the injection of the
89 silicone on a semi-liquid state into the nozzle, once the needle has been removed. After a
90 few hours, the silicone becomes solid and can be extracted, maintaining the internal

91 geometry of the sac and the orifices. The mold is later inspected using a Scanning Electron
92 Microscope, determining the corresponding nozzle dimensions.

93 An example of the pictures obtained through this process can be seen in Figure 1, while
94 more details on the experimental technique are available in [42]. Finally, the final
95 geometrical values of the nozzle used for the study can be seen in Table 1. In this table,
96 R_a and R_b are the rounding radii at the orifice inlet in the upper and lower side of the
97 orifice, respectively; D_i , D_o and D_m are the diameters in the inlet, outlet and middle
98 sections of the orifices; and k -factor is a parameter related to the nozzle orifice conicity,
99 defined as:

$$k - factor = \frac{D_i[\mu m] - D_o[\mu m]}{10} \quad (1)$$

100 Since the nozzle orifices are significantly convergent (as it can be seen from its high value
101 of k -factor), low probability of cavitation formation inside the nozzle is expected [43],
102 [44]. Nevertheless, some cavitation could appear when very high injection pressures are
103 used. This will be further analyzed in Section 4.

104 **2.2. Fuel properties characterization**

105 As a first step, the main physical properties of the fuel have been measured under a wide
106 range of pressure and temperature conditions. In particular, a standard European winter
107 diesel fuel has been used. Density measurements were performed on a hydrometer, based
108 on the ASTM D1298 procedure, while a standard viscometer was used to characterize the
109 fuel viscosity. Finally, a custom-made facility was constructed to characterize the speed
110 of sound. This facility was based on a standard common-rail system, onto which a long
111 tube has been installed between the rail and the injector. On that line, two high-speed

112 piezoelectric pressure transducers have been installed at two different positions. Once the
 113 injector is commanded and the injection event takes place, a pressure wave is generated
 114 inside the system. Knowing the distance between these two transducers, it is possible to
 115 characterize the speed of sound by measuring the time lapse that the pressure perturbation
 116 takes to travel to one sensor to another. More information about the experimental setup
 117 can be seen in [45].

118 Figure 2 shows the results from the fuel characterization for a range of 0.1-300 MPa in
 119 pressure and 300-400 K in temperature, which are representative of the usage of diesel
 120 fuel in advanced common rail systems. These data have been correlated as a function of
 121 pressure and temperature, finding the following relationships:

$$\rho_f = 826.5 - 1.0217(T - 298) + 1.251 \cdot 10^{-3}(T - 298)^2 + 0.6035(P - 0.1) - 8.27 \cdot 10^{-4}(P - 0.1)^2 + 1.44 \cdot 10^{-3}(P - 0.1)(T - 298) \quad (2)$$

$$\mu_f = 10^{-3} \mu_0 \cdot 10^{\left[(-1.48 + 5.86 \mu_0^{0.181}) \left(\frac{P - 0.1013}{1000}\right)\right]} \quad (3)$$

$$\mu_0 = 3.2158 \cdot \exp[0.0263(T - 298)] \quad (4)$$

$$a_f = 1350.6 - 3.1485(T - 298) + 4.4928(P - 0.1) - 6.96 \cdot 10^{-3}(P - 0.1)^2 + 7.4 \cdot 10^{-3}(P - 0.1)(T - 298) \quad (5)$$

122 Where ρ_f is the fuel density in kg/m³, μ_f is the fuel dynamic viscosity in Pa·s, μ_0 is the fuel
 123 dynamic viscosity at 0.1 MPa of pressure, a_f is the speed of sound of the fuel in m/s, P is
 124 the fuel pressure in MPa and T is the fuel temperature in K.

125

126

127 **2.3. Injection rate meter**

128 An IAV injection rate meter has allowed to determine the instantaneous mass flow rate
129 delivered by the injector at different boundary conditions, summarized in Table 2. The
130 technique is based on the Bosch method [46], which relates the instantaneous injected
131 quantity to the pressure increase on a tube placed at the injector outlet. More details on
132 the experimental arrangement and postprocessing procedure can be found in [47].

133 Figure 3 shows an example of the results obtained for a particular case of 180 MPa
134 injection pressure (P_{inj}), 2 ms of energizing time (ET) and different levels of backpressure
135 (P_b). The curve represents the instantaneous mass flow injected by the combination of the
136 5 nozzle orifices. As it is usual for the high injection pressure cases, the effect of the
137 backpressure is only appreciable on the steady-state phase of the injection event. During
138 this region, the mass flow rate through the nozzle corresponds to the following
139 expression:

$$\dot{m} = C_d \rho_f A_o u_b = C_d A_o \sqrt{2(P_{inj} - P_b)} \rho_f \quad (6)$$

140 Where \dot{m} is the mass flow rate through the nozzle, C_d is the discharge coefficient of the
141 nozzle, A_o is the geometrical outlet area of the nozzle orifices and u_b is the theoretical
142 nozzle outlet velocity according to Bernoulli's equation. According to this expression,
143 and as it can be seen in Figure 3, higher backpressure values correspond to lower
144 stationary mass flow rates.

145 **2.4. Momentum flux test rig**

146 A dedicated test rig has allowed to obtain the momentum of the sprays produced by the
147 injector. In this rig, the fuel is injected on chamber filled with an inert pressure gas

148 (nitrogen in this case). A compound of a piezoelectric pressure transducer and a target is
149 placed perpendicular to one of the orifices of the fuel injector, at a distance of 5 mm. The
150 transducer is properly calibrated so that it can measure the impact force of the spray into
151 the target, which is then transmitted to the transducer. More details of the technique can
152 be found in [48]. The experimental matrix for the momentum flux measurements, which
153 is a subset of the one already seen for the injection rate meter, is available in Table 3. In
154 this case, the maximum backpressure was limited to 7 MPa due to structural limitations
155 of the test rig.

156 Figure 4 shows a schematic of the momentum flux experimental arrangement, together
157 with an example of the results again for the $P_{inj}=180$ MPa case. The results seen in the
158 figure are an average of the data coming from the 5 nozzle orifices.

159

160 **3. NUMERICAL SETUP**

161 Internal nozzle flow simulations have been carried out using a single-phase isothermal
162 flow solver in ANSYS ® Fluent ® v.17 [49]. Regarding the turbulence model, Re-
163 Normalization Group (RNG) k- ϵ model has been selected based on previous internal flow
164 simulation experiences [41], [50]. The geometry has been simplified to a 72° sector-mesh,
165 corresponding to a single nozzle orifice, in order to minimize the computational effort.
166 The mean orifice dimensions included in Table 2 have been used for this purpose.

167 Figure 5 shows the computational domain with a detail of the mesh structure in the orifice.
168 Constant pressure boundary condition is selected for the Inlet and Outlet boundary
169 conditions, with values equal to the experimental data at the fuel injector inlet and outlet

170 during the injection rate experimental campaign. Non-slip boundary condition is used for
171 the nozzle and needle walls.

172 Figure 6 shows the results for a mesh sensitivity study and the comparison between first
173 and second order numerical schemes. This has been performed for an injection pressure
174 of 130 MPa and a backpressure of 7 MPa. The fuel properties (density and viscosity) have
175 been considered constant along the whole computational domain (incompressible
176 solution). From Figure 6, it can be observed that second order schemes reach the mesh
177 independence for a relatively small number of cells (#208000), while first order numerical
178 schemes do not show mesh convergence for significantly higher number of cells
179 (#272000). Thus, in order to minimize the computational effort of the simulations, second
180 order schemes with the 208000 cells configuration have been selected for the study. This
181 configuration leads to an overestimation in the experimental mass flow rate of
182 approximately 5.5%.

183 Using the previously determined mesh characteristics, all cases presented in Table 2 have
184 been run on a single processor Intel (R) Core (TM) i5-4460 CPU @ 3.20 GHz. The
185 simulations have been run on a steady-state solver, using two main convergence criteria:
186 first, all the residuals must be below $5 \cdot 10^{-5}$; additionally, the average velocity at the
187 nozzle orifice outlet must reach stationary conditions with 1% tolerance. The simulations
188 are initialized with injection pressure and zero velocity in the internal fluid domain. Doing
189 so, and for the particular case of an injection pressure of 130 MPa and a backpressure of
190 5 MPa, the incompressible solver reaches convergence after 1778 iterations, leading to a
191 total CPU time of 2421.7 seconds. For the same conditions, the compressible approach
192 takes 1860 iterations and 3059.9 seconds to converge, which represents an increase of the

193 computational effort of approximately 26%. Similar results on a relative basis are
194 obtained for other operating conditions.”

195

196 **4. INTERNAL FLOW SIMULATION RESULTS**

197 In the current section, internal nozzle flow simulation results will be analyzed comparing
198 two different strategies. First, the fuel properties are considered constant for the whole
199 computational domain. For this purpose, density and viscosity are calculated at the
200 backpressure condition. Then, the equations described in Section 2 for the density,
201 viscosity and speed of sound of the fuel are introduced into the solver by means of user-
202 defined functions, in order to account for the fuel compressibility. For both cases, the
203 flow is considered isothermal with a temperature level of 298 K, which is the value
204 existing at the fuel injector inlet during the experimental injection rate and momentum
205 flux measurements.

206 Figure 7 shows an example of the density and viscosity fields inside the nozzle for the
207 compressible configuration. As a consequence of the pressure evolution inside the nozzle,
208 which will be later analyzed in Figure 8, the compressible solver estimates a variation of
209 around 100 kg/m^3 in density and of around $8 \cdot 10^{-2} \text{ kg/m}\cdot\text{s}$ in dynamic viscosity along the
210 computational domain. These variations have a double impact: on the one hand, the
211 variations in the fuel properties are expected to induce significant changes in the nozzle
212 outlet velocity and mass flow rate compared to the incompressible solution, as it will
213 analyzed in the next paragraphs; on the other hand, the variations of viscosity affect the
214 local Reynolds number, with consequences in the turbulent flow characteristics.

215 Figure 8 shows the comparison of pressure and velocity fields for the compressible and
216 incompressible solvers for the same condition analyzed in Figure 7. One of the first things
217 that can be highlighted is that even for this very high injection pressure, the minimum
218 computed pressure values inside the nozzle orifices (around 0.5 MPa) are always higher
219 than the fuel saturation pressure. This means that this particular geometry would not
220 produce any cavitation thanks to the combination of high conicity and relatively high
221 rounding radii. Another significant difference is seen in the velocity fields. In the
222 compressible solution, higher fuel viscosity values are observed compared to the
223 incompressible approach, where the viscosity is calculated at backpressure conditions.
224 For this reason, higher viscous dissipation appears, leading to lower velocities. This can
225 be easily perceived looking at the maximum velocity along the computational domain,
226 which is around 58 m/s lower for the compressible case. Additionally, both pressure and
227 velocity contours show smoother transitions along the computational domain when
228 including the flow compressibility effects.

229 The impact of the variation of the properties inside the nozzle over the hydraulic behavior
230 of the nozzle can be clearly observed in Figure 9. This figure compares the experimental
231 mass flow at the nozzle outlet with the simulation results obtained with and without the
232 compressibility equations enabled. In the case of the experimental results, the data
233 corresponds to a time average of the injection rate during its steady-state phase, where
234 the mass flow is not affected by the needle position, as it was introduced in Section 2. In
235 these results, the simulations tend to overestimate the mass flow for all conditions. At
236 relatively low injection pressures (30 MPa), the solution given by the compressible and
237 incompressible approaches are relatively similar, since the range of variation of the fluid
238 properties is moderate. However, it is appreciable that the compressible solution is closer

239 to the experimental values, as it is more capable of representing the flow physics. As the
240 injection pressure increases, the compressible and incompressible solutions diverge,
241 reaching a maximum difference of approximately 5% in mass flow at the maximum
242 injection pressure tested (250 MPa).

243 Figure 10 shows the evolution of the discharge coefficient against the square root of the
244 pressure drop for both compressible and incompressible flow, together with the
245 experimental values. For all of them it can be seen how the discharge coefficient is highly
246 dependent on the pressure drop along the nozzle at low $\Delta P^{1/2}$ conditions, while the
247 dependence is much smaller as the pressure drop increases. This behavior is due to the
248 impact of the flow regime on the discharge coefficient. At low injection pressures, flow
249 velocities are moderate and the flow is in transitional conditions between laminar and
250 turbulent, for which the discharge coefficient is highly sensitive to the Reynolds number.
251 As the injection pressure increase, so does the velocity, the flow regime becomes fully
252 turbulent and the discharge coefficient is independent on the Reynolds number. Similar
253 behavior has been repeatedly found in the literature for different orifice geometries [51]–
254 [53]

255 Regarding the effect of compressibility, at low injection pressures both approaches clearly
256 overestimate the discharge coefficient, although the compressible solution gives better
257 results. The relatively high difference between model and experiments at these conditions
258 may be due to uncertainties in aspects such as the nozzle geometry or the turbulence
259 model. As the injection pressure increases, the importance of the flow compressibility
260 ramps up, the compressible and incompressible solutions diverge, and it is clearly seen
261 how the compressible flow solver is more capable to reproduce the experimental data,

262 while the incompressible solver maintains a deviation of approximately 0.08 in the
263 absolute value of the discharge coefficient.

264 Figure 11 plots the hydraulic performance of the nozzle in terms of its effective outlet
265 velocity, calculated as the ratio between the momentum flux and the mass flow. For the
266 experiments, the time-averaged values at the steady-state phases of the injection rate and
267 momentum flux curves are considered. For the CFD calculations, the mass flow and
268 momentum flux values are integrated in the nozzle outlet section. It can be observed that
269 the incompressible solution overestimates again the outlet velocity, while the values
270 obtained using the compressible flow approach are very similar to the experiments for all
271 the conditions tested.

272 Finally, Figures 12 and 13 show the mass flow and effective outlet velocity results
273 expressed as the percentage deviation to the experimental data. In both cases, it can be
274 seen how this deviation tends to reduce when increasing the injection pressure.
275 Comparing the two simulation approaches, the compressible flow solution is around 5%
276 closer to the experiments in terms of mass flow. This fact points out the importance of an
277 accurate reproduction of the fluid properties when trying to reproduce the hydraulic
278 behavior of a nozzle through simulations. Regarding the effective velocity, the results
279 pass from around 3% of overestimation in the high-pressure range for the incompressible
280 simulation to a 1% underestimation in the case of the compressible solution, while the
281 deviations are significantly higher for the constant-properties approach.

282 **5. CONCLUSIONS.**

283 In the current paper, an investigation of the compressibility effects on diesel nozzle
284 internal flow simulations has been performed. First, the fluid density, viscosity and speed

285 of sound have been characterized as a function of pressure and temperature on a wide
286 range of boundary conditions. Correlations of these properties have been estimated and
287 then implemented on ANSYS ® Fluent ® v.17. Later, the hydraulic behavior of a 5-
288 orifices convergent nozzle has been characterized by means of mass flow rate and
289 momentum flux experimental tests, exploring values of injection pressure up to 250 MPa.
290 From these results, the evolution of the mass flow rate, momentum flux and effective
291 outlet velocity at maximum lift conditions have been extracted. Then, these values are
292 compared to steady-state CFD simulations at two conditions:

- 293 - Incompressible flow: constant fluid properties.
- 294 - Compressible flow: fluid properties locally computed as a function of the flow
295 pressure conditions.

296 The results from the compressible flow simulations show a variation of around 100 kg/m^3
297 in density and of around $8 \cdot 10^{-2} \text{ kg/m}\cdot\text{s}$ in dynamic viscosity along the nozzle geometry.
298 This implies a significant reduction of the uncertainties related to internal nozzle flow
299 simulations without significant impact in the computational effort. In particular, the
300 accuracy in the prediction of the mass flow rate improves around 5% when using the
301 compressible flow approach. Other flow characteristics such as the momentum flux and
302 the effective outlet velocity, which are key inputs for spray models, also show a
303 significant improvement in accuracy.

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 489 performance of diesel injection nozzles,” *Energy Convers. Manag.*, vol. 105, pp.
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Tables

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Orifice	Ra [μm]	Rb [μm]	Di [μm]	Dm [μm]	Do [μm]	k-factor [-]	Length [μm]
1	23	13	141	129	123	1.7	703
2	27	23	145	131	127	1.8	704
3	20	24	145	132	126	1.9	707
4	36	20	141	128	122	1.8	726
5	38	28	145	130	123	2.2	737

Mean Values	29±8	22 ±6	143 ±2.3	130 ±1.4	124 ±2.2	1.9 ±0.2	715 ±15
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497 **Table 1. Nozzle geometric characteristics**

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Injection Pressures [MPa]	Back-Pressures [MPa]								
30	0.5	1	3	5	7	9	11	13	15
80		1	3	5	7	9	11	13	15
130		1	3	5	7	9	11	13	15
180		1	3	5	7	9	11	13	15
250		1	3	5	7	9	11	13	15

500 **Table 2. Text matrix for mass flow rate measurements.**

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Injection Pressures [MPa]	Back-Pressures [MPa]					
30	0.5	1	3	5	7	
80		1	3	5	7	
130		1	3	5	7	
180		1	3	5	7	
250		1	3	5	7	

503 **Table 3. Text matrix for Momentum flux measurements.**

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Figure captions.

508 Figure 1. Nozzle geometry determination.

509 Figure 2. Winter diesel properties as a function of pressure and temperature.

510 Figure 3. Mass flow rates at $P_{inj}= 180$ MPa and all back-pressures.

511 Figure 4. Momentum flux at $P_{inj}= 180$ MPa and all back-pressures.

512 Figure 5. Details of nozzle mesh.

513 Figure 6. Mesh sensibility study for first and second order schemes.

514 Figure 7. Fields of density and viscosity obtained from CFD simulations (compressible
515 approach) for an injection pressure of 250 MPa and backpressure of 5 MPa

516 Figure 8: Fields of pressure and velocity for compressible and incompressible solutions.
517 $P_{inj} = 250$ MPa, $P_b = 5$ MPa.

518 Figure 9. Experimental mass flow results compared to those of CFD calculations for
519 incompressible and compressible approaches.

520 Figure 10. Experimental discharge coefficient results compared to those of CFD
521 calculations for incompressible and compressible approaches.

522 Figure 11. Experimental effective injection velocity results compared to those of CFD
523 calculations for incompressible and compressible approaches.

524 Figure 12. Mass flow deviation among experimental and modelled (incompressible and
525 compressible approaches)

526 Figure 13. Velocity deviation among experimental and modelled (incompressible and
527 compressible approaches)

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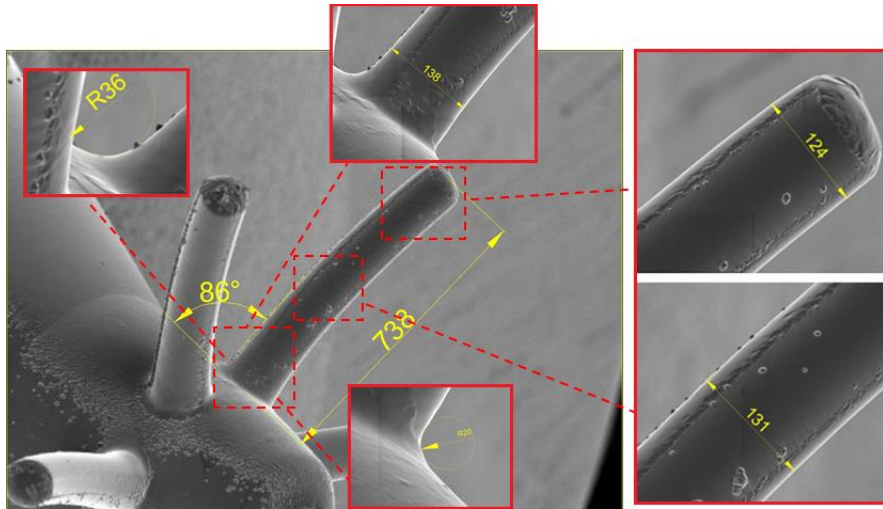
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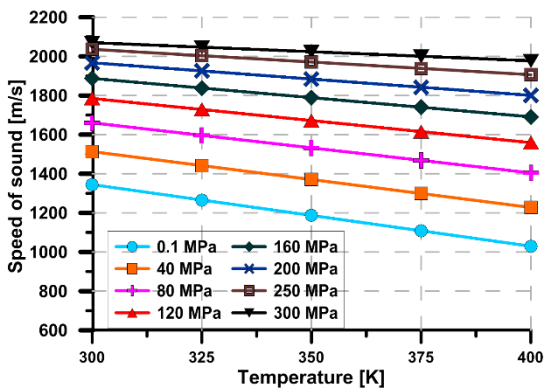
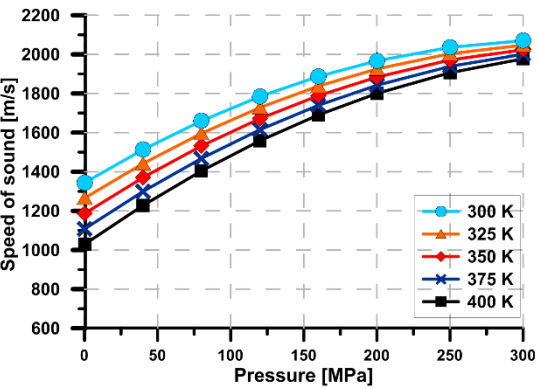
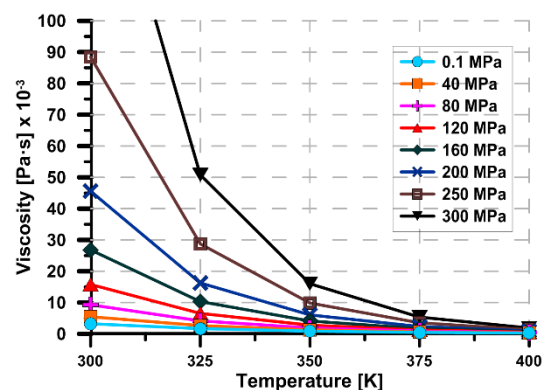
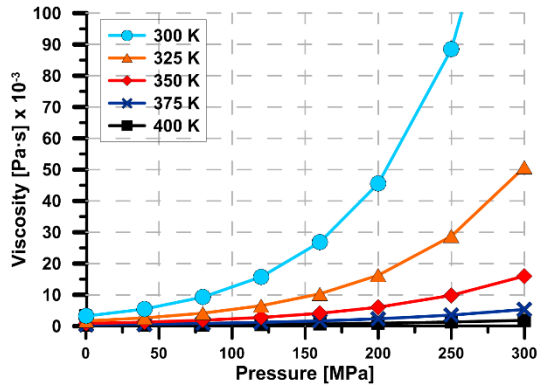
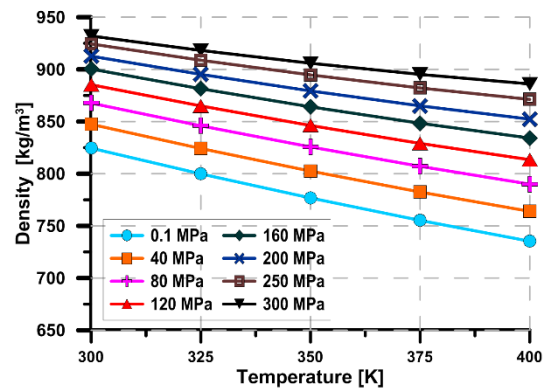
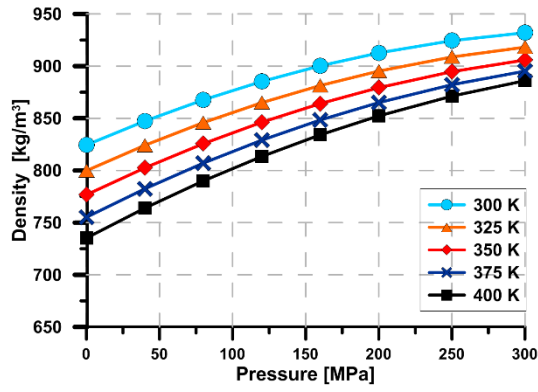


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Figure 1. Nozzle geometry determination.

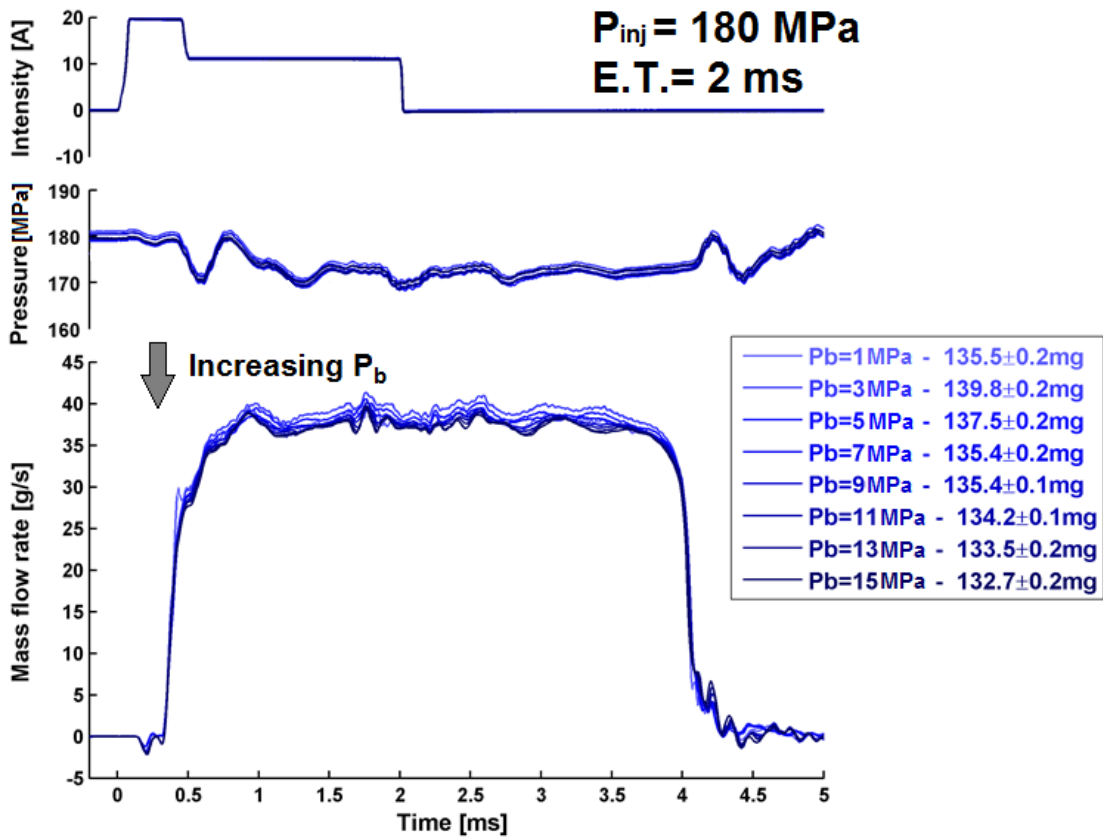


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Figure 2. Winter diesel properties as a function of pressure and temperature.

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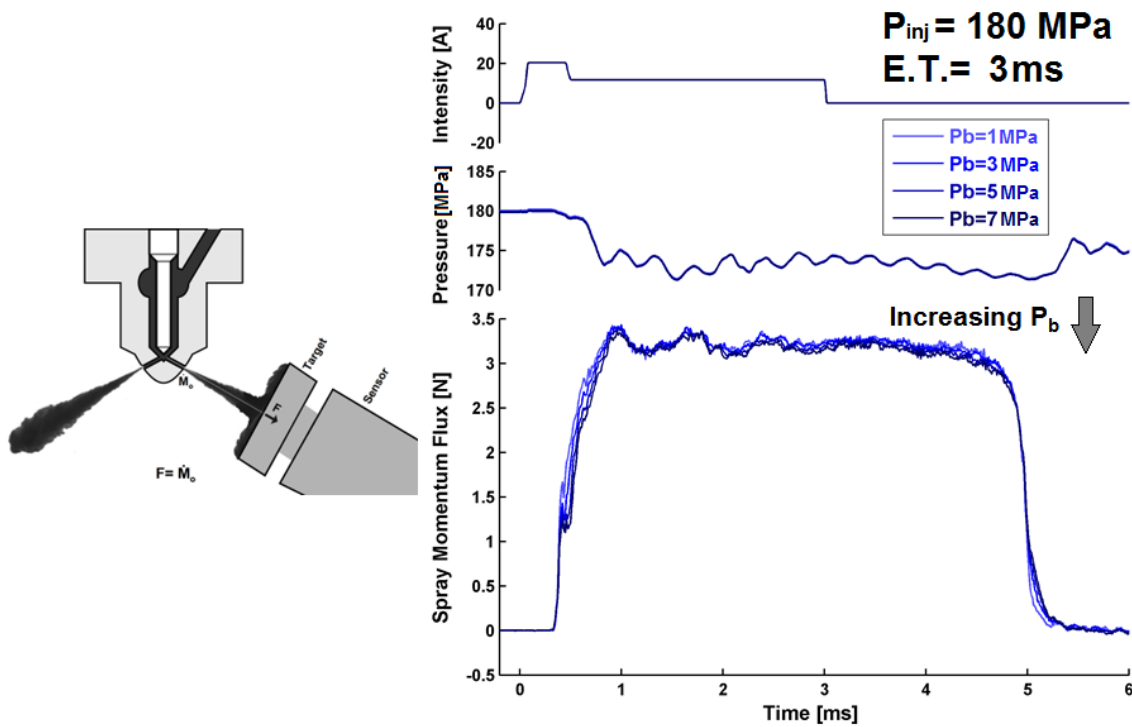


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Figure 3. Mass flow rates at $P_{inj} = 180 \text{ MPa}$ and all back-pressures.

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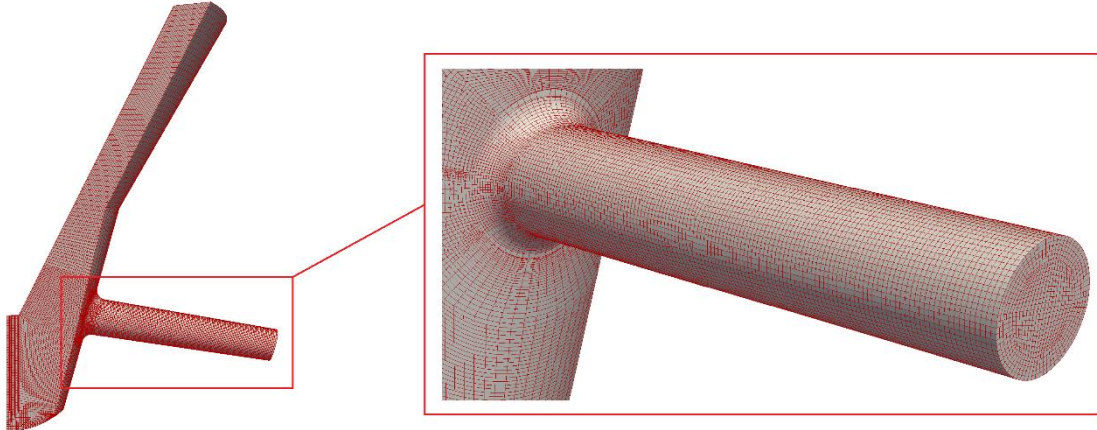
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Figure 4. Momentum flux at $P_{inj} = 180 \text{ MPa}$ and all back-pressures.

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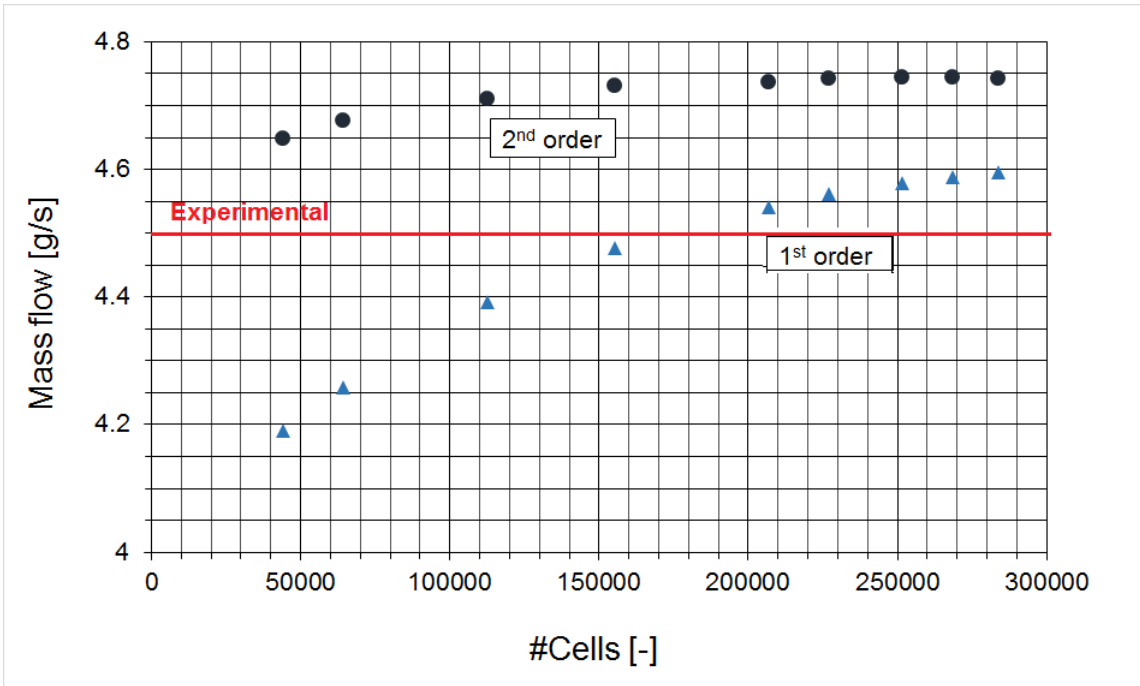


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Figure 5. Details of nozzle mesh.

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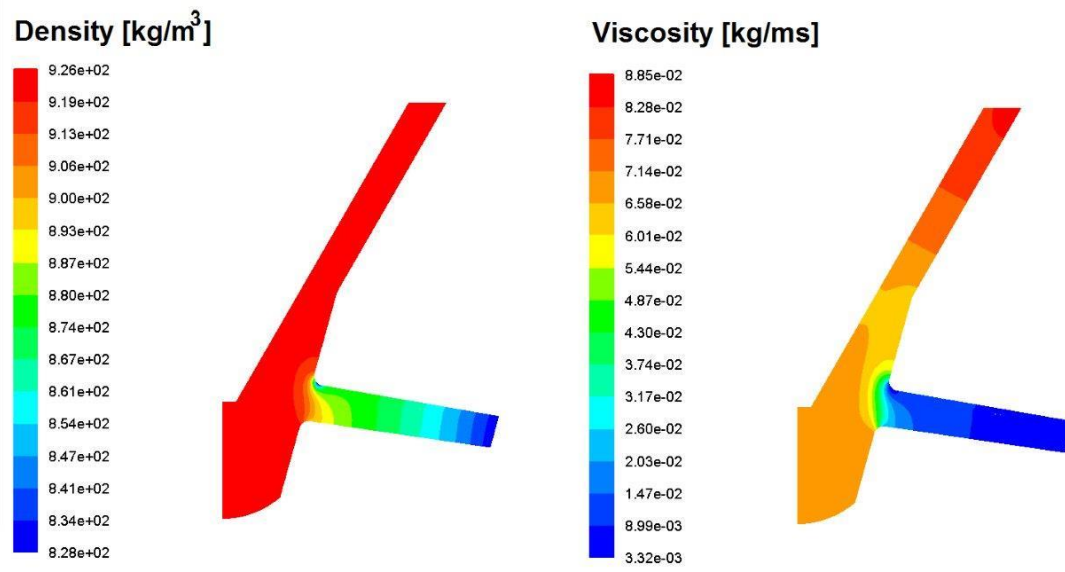
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Figure 6. Mesh sensitivity study for first and second order schemes. $P_{inj} = 130$ MPa, $P_b = 5$ MPa

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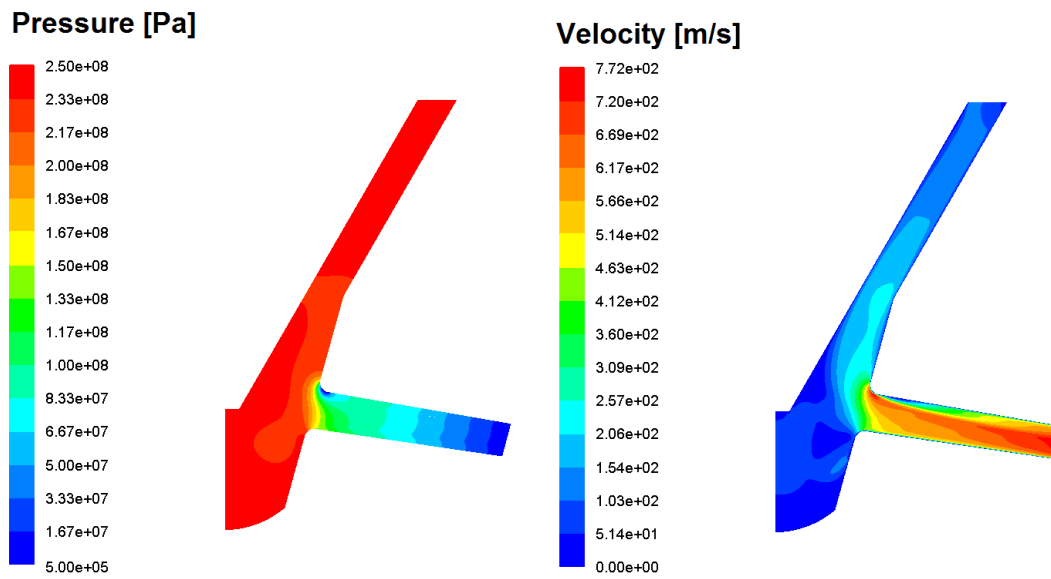


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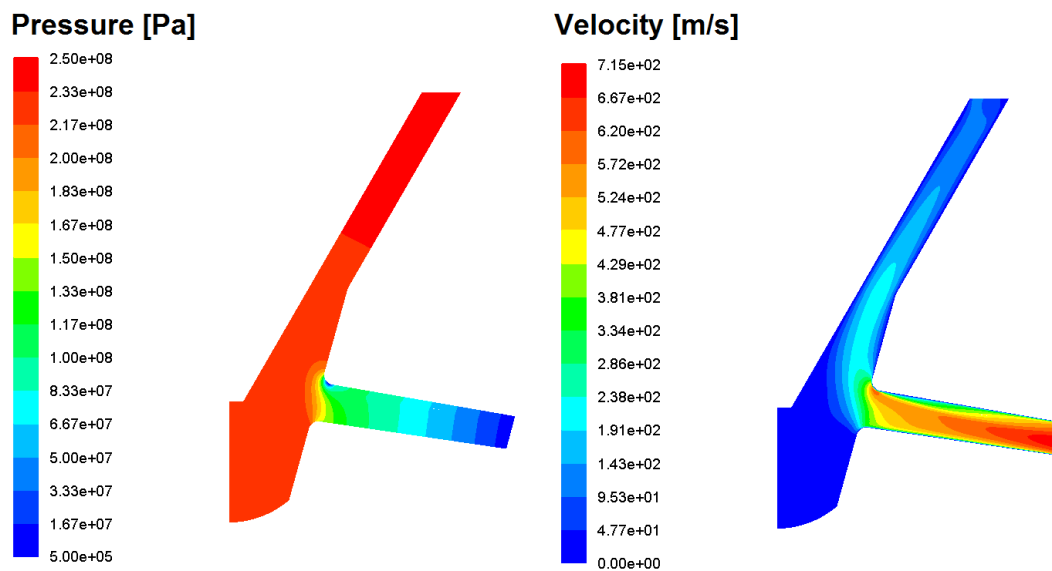
561 Figure 7. Fields of density and viscosity obtained from CFD simulations (compressible
562 approach) for an injection pressure of 250 MPa and backpressure of 5 MPa

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INCOMPRESSIBLE APPROACH



COMPRESSIBLE APPROACH

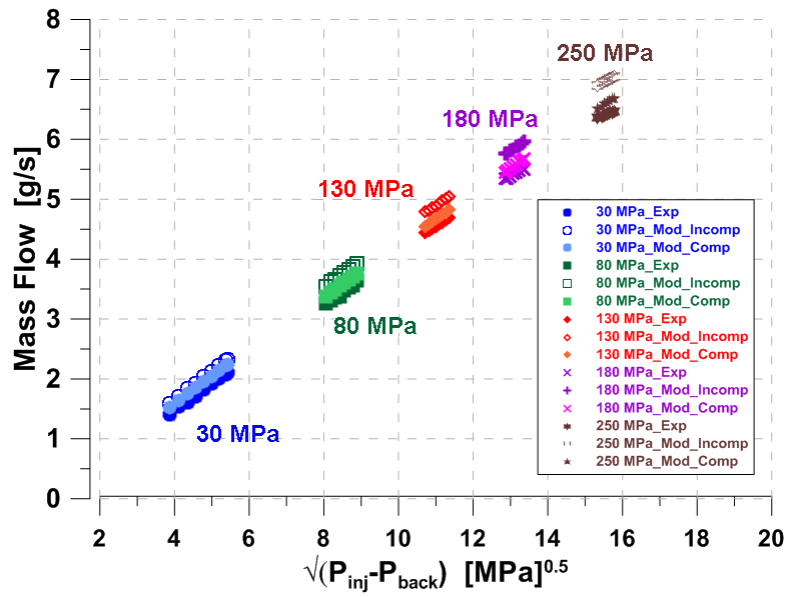


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565 Figure 8: Fields of pressure and velocity for compressible and incompressible solutions.
566 $P_{inj} = 250$ MPa, $P_b = 5$ MPa.

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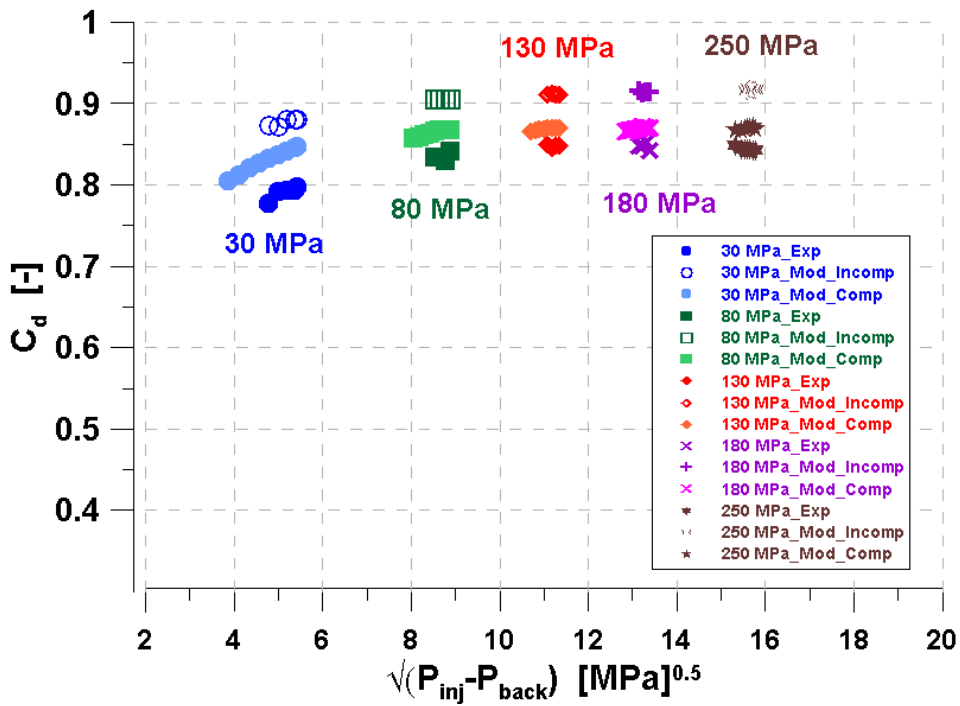
570 Figure 9. Experimental mass flow results compared to those of CFD calculations for
 571 incompressible and compressible approaches.

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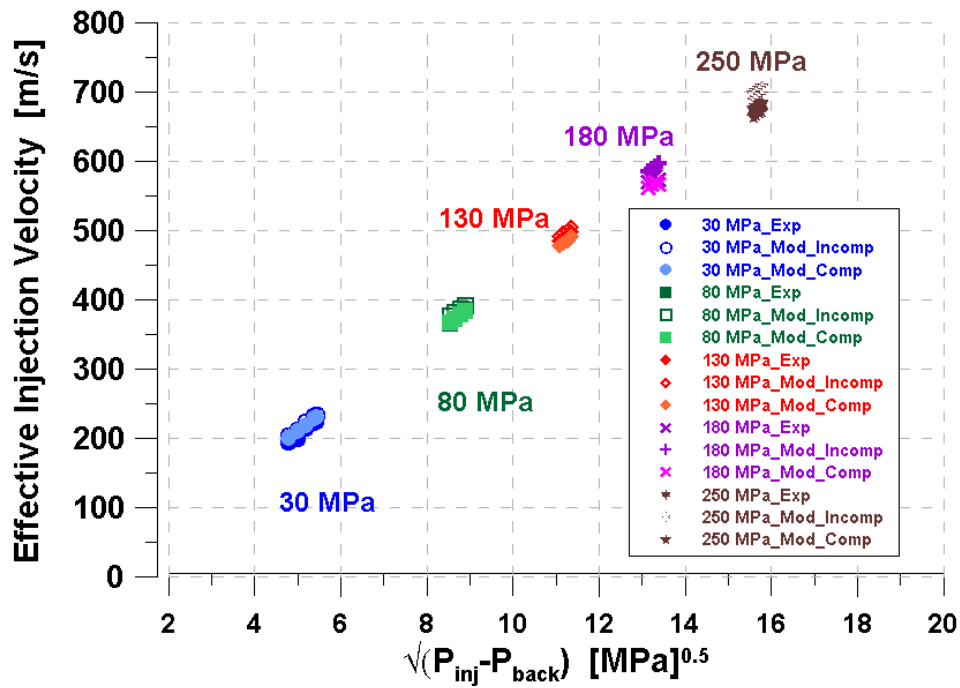
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577 Figure 10. Experimental discharge coefficient results compared to those of CFD
 578 calculations for incompressible and compressible approaches.

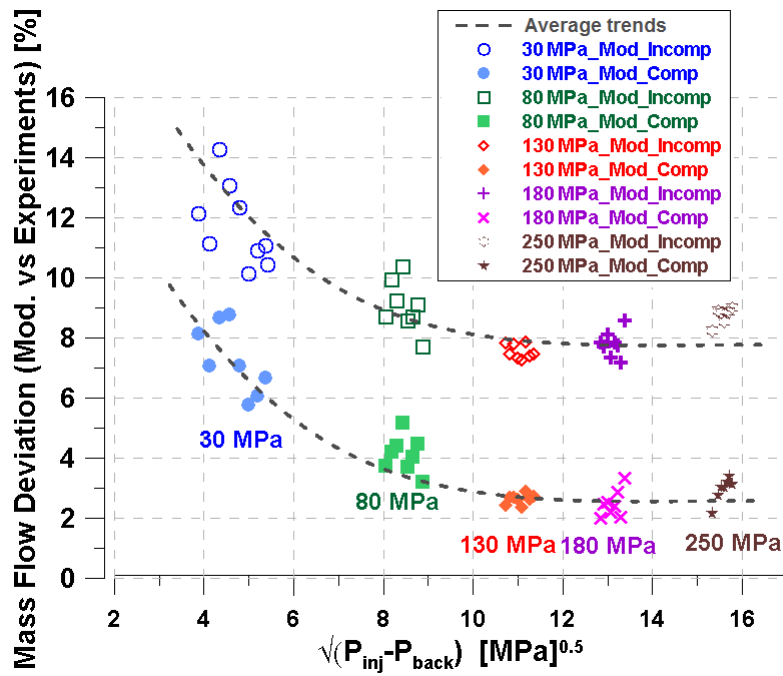
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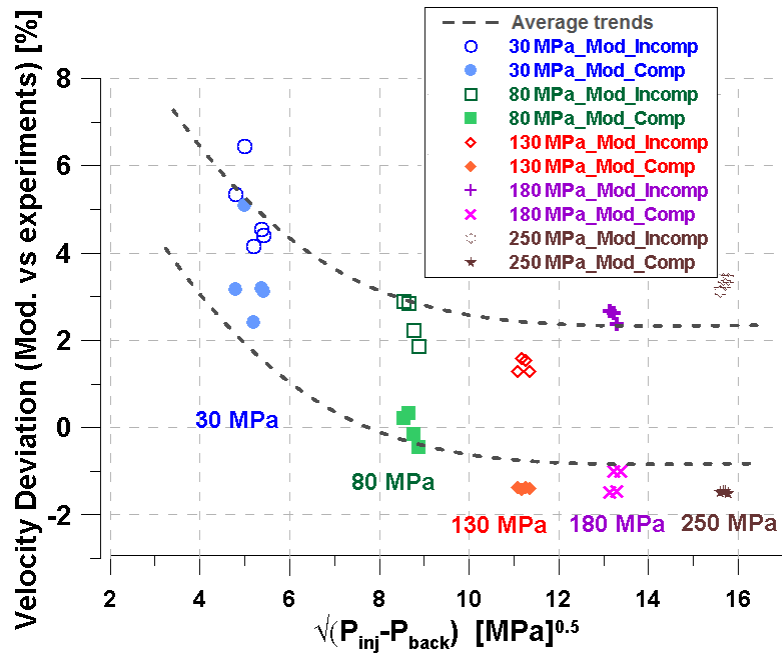
581 Figure 11. Experimental effective injection velocity results compared to those of CFD
582 calculations for incompressible and compressible approaches.

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585 Figure 12. Mass flow deviation among experimental and modelled (incompressible and
586 compressible approaches)



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588 Figure 13. Velocity deviation among experimental and modelled (incompressible and
 589 compressible approaches)

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